4. The turbine engine component of claim **1**, wherein said at least one turbine engine component comprises said bypass fan duct having an interior surface comprising at least one aperture.

5. The turbine engine component of claim **4**, wherein in addition to said bypass fan duct said at least one turbine engine component comprises any one of the following: a turbine exhaust case, a turbine exhaust nozzle, and a bypass air valve support.

6. The turbine engine component of claim **5**, wherein said turbine engine component comprises said turbine exhaust case having an exterior surface with at least one aperture.

7. The turbine engine component of claim **5**, wherein said turbine engine component comprises said turbine exhaust nozzle having an exterior surface with at least one aperture.

8. The turbine engine component of claim **5**, wherein said turbine engine component comprises said bypass air valve support having an exterior surface with at least one aperture.

9. A process for controlling fan stream flow bypass of a turbofan engine, comprising:

providing a turbine engine component having a surface including at least one aperture, and located from between a bypass fan duct and a turbine exhaust nozzle of the turbofan engine;

providing a bypass air valve having a surface including at least one aperture and at least one impermeable region, and concentrically disposed about said turbine engine component and parallel to a centerline of the turbofan engine;

introducing a fan exhaust stream flow into the turbofan engine;

actuating said bypass air valve to substantially align said at least one aperture of said bypass air valve with said at least one aperture of said turbine engine component; and

permitting flow transfer by substantially aligning said at least one apertures and transferring said fan exhaust stream flow into a turbine exhaust stream flow of the turbofan engine at a flow transfer location.

10. The process of claim **9**, wherein actuating said bypass air valve comprises rotating said bypass air valve about said turbine engine component.

11. The process of claim **9**, wherein actuating said bypass air valve comprises moving said bypass air valve in a forward direction or a backward direction parallel to the centerline along the turbine engine component.

12. The process of claim **9**, wherein permitting said flow transfer further comprises effectively increasing a fan exhaust stream nozzle area within said fan exhaust stream flow without altering a fan exhaust nozzle and effectively decreasing a turbine exhaust stream nozzle area within said turbine exhaust stream flow without altering a turbine exhaust nozzle.

13. The process of claim **9**, further comprising the steps of:

actuating said bypass air valve to substantially align said at least one impermeable region with said at least one aperture of said turbine engine component; and

ceasing the transfer of said fan exhaust stream flow into said turbine exhaust stream flow.

14. The process of claim **13**, wherein ceasing the transfer further comprises effectively decreasing a fan exhaust stream nozzle area within said fan exhaust stream flow without altering a fan exhaust nozzle and effectively increasing a turbine exhaust stream nozzle area within said turbine exhaust stream flow without altering a turbine exhaust nozzle.

15. The process of claim **9**, further comprising increasing a design bypass ratio to increase the ratio of a fan exhaust stream pressure to a turbine exhaust stream pressure.

16. The process of claim **9**, further comprising selecting a flow transfer location exhibiting an increased Mach Number to provide an increased ratio of a total pressure to a static pressure at said flow transfer location.

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